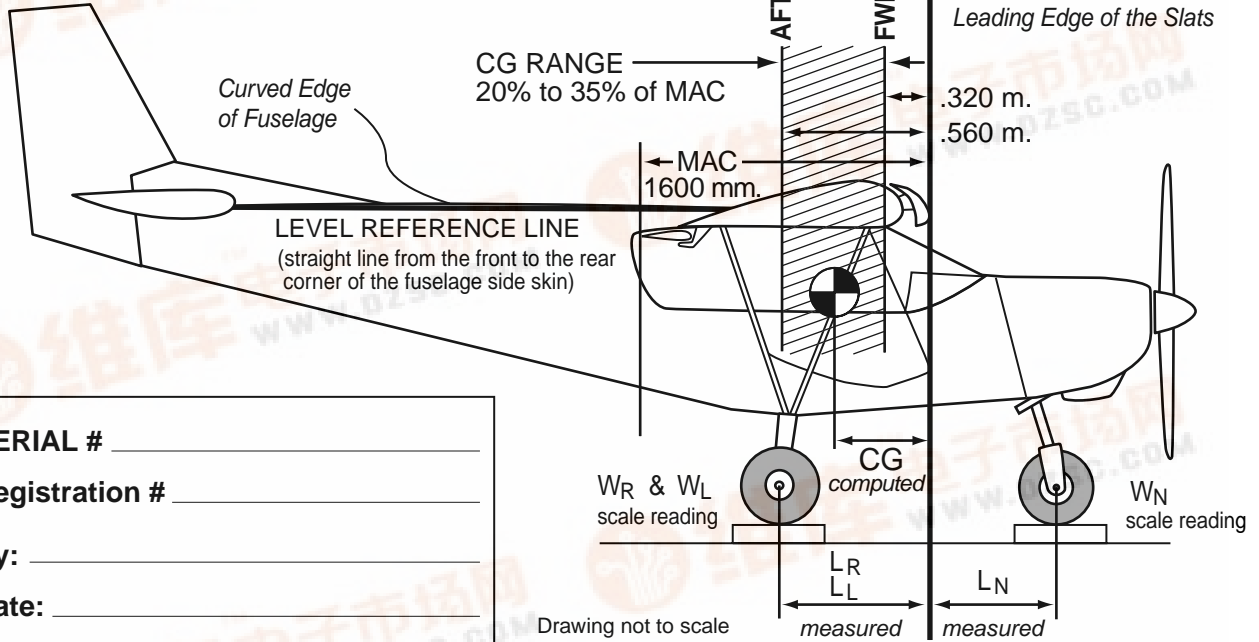


STOL CH 801 WEIGHT & BALANCE REPORT

Forward & Aft C.G. Check



SERIAL # _____
Registration # _____
By: _____
Date: _____

Drawing not to scale

ITEM	WEIGHT (pounds)	ARM (meters)	MOMENT
AIRCRAFT EMPTY CG	RIGHT MAIN WHEEL	$W_R =$	$L_R =$
	LEFT MAIN WHEEL	$W_L =$	$L_L =$
	NOSE WHEEL	$W_N =$	$L_N =$ - negative arm
	COMPUTED CG EMPTY	Empty Weight:	CG=
		Arm to Datum	Aircraft Moment

		MOMENT - Forward	MOMENT - Rear
PILOT		.50 *	
PASSENGER		.50 *	
REAR SEAT		1.4	
FUEL: WING TANKS L & R: ____ Gal. each		.60	
FUEL: EXTENDED RANGE		.60	
TOTAL	$W_{FRD} =$	$M_{FRD} =$	$M_{AFT} =$
	$W_{AFT} =$		
Gross Weight:	Take-Off Weight:	$CG_{FRD} =$	$CG_{AFT} =$

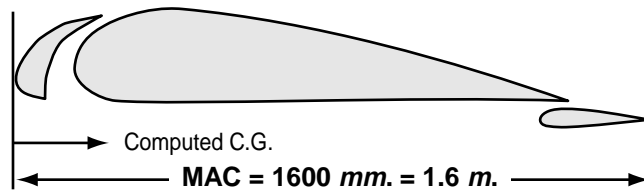
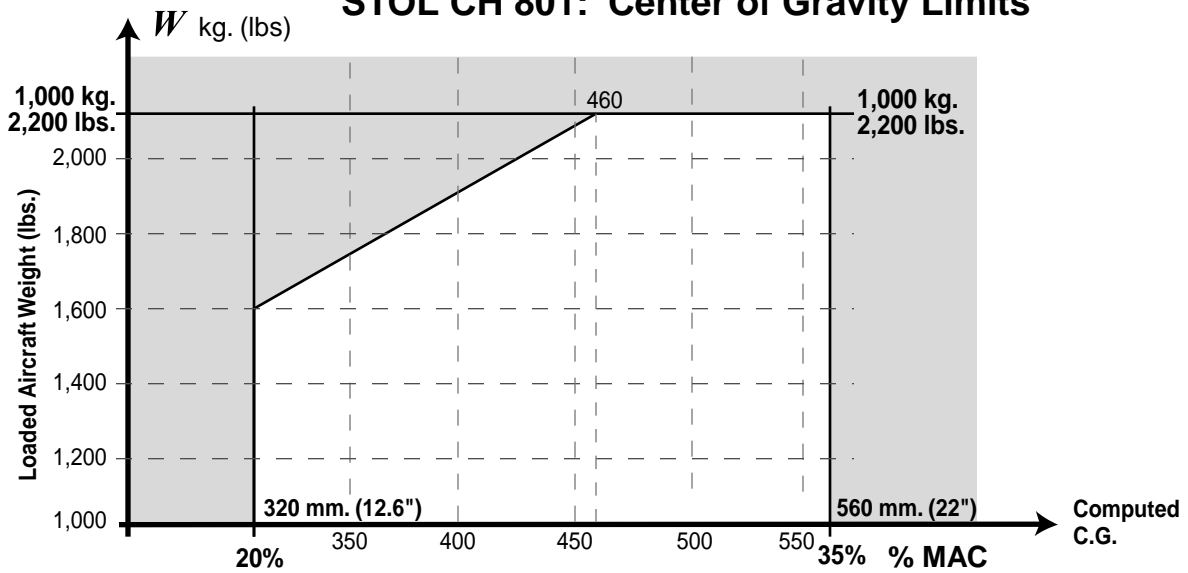
* Depends on seat position & individuals' weight distribution.

CG Range: From .320 m. to .560 m.

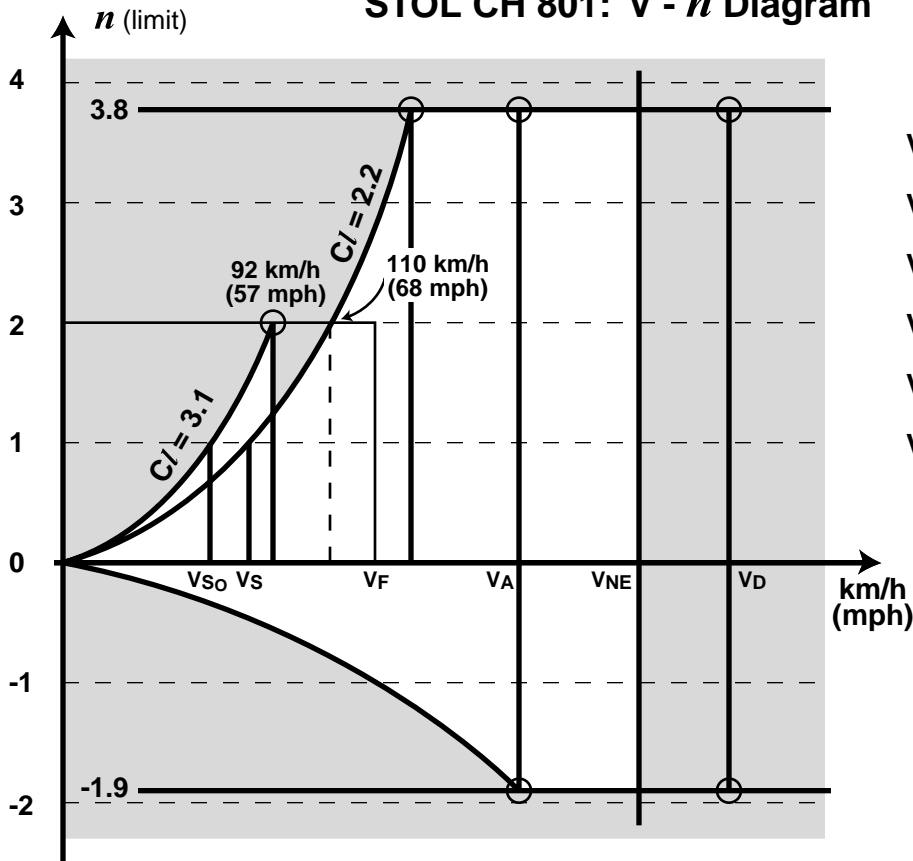
$$\text{Center of Gravity (CG)} = \frac{\text{Total Moment}}{\text{Total Weight}}$$



STOL CH 801: Center of Gravity Limits



STOL CH 801: V - n Diagram



- $V_{So} = 65 \text{ km/h (40 mph)}$
- $V_s = 78 \text{ km/h (48 mph)}$
- $V_F = 130 \text{ km/h (80 mph)}$
- $V_A = V_C = 193 \text{ km/h (120 mph)}$
- $V_{NE} = 240 \text{ km/h (150 mph)}$
- $V_D = 270 \text{ km/h (168 mph)}$

